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CSM BATTERY CHARGER MODEL



Guidance and Performance Branch

MISSION PLANNING AND ANALYSIS DIVISION



MANNED SPACECRAFT CENTER
HOUSTON, TEXAS

(NASA-TM-X-69690) MODEL (NASA) 41 p CSM BATTERY CHARGER

N74-70524

Unclas 00/99 16316

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By James McClellan and R. E. Stokes Guidance and Performance Branch

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MISSION PLANNING AND ANALYSIS DIVISION NATIONAL AERONAUTICS AND SPACE ADMINISTRATION MANNED SPACECRAFT CENTER HOUSTON, TEXAS

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1.0 SUMMARY AND INTRODUCTION

The purpose of this note is to present the design concept, assumptions, logic flow chart, program listing, and a sample case of the CSM battery charger model. Basically, the model simulates the battery charger during the charge cycle of the entry and postlanding batteries. The model is to be used in the CSM SEENA program so that more accurate CSM entry and postlanding battery power profiles may be generated. The model will also be used to simulate the CSM battery charger in the AAP EPS program.

2.0 SYMBOLS

A, B, C, E, F tables of data for charge rate curve

AF degradation factor

A-h ampere hours

AHD amp-hours discharged from Batt I

AHS(I) amp-hours present in Batt I

AHST present amp-hour status

AMP(I) amps into Batt I

Batt I CSM entry and postlanding batteries

CALC intermediate variable used for comparison only

CP intermediate variable (%) read from C table and used to calculate C (C = CP * B)

CSTS CST1 + CST2

CST1 area under charge rate curve from 0 to E

CST2 area under charge rate curve from E to F

DELCH delta charge, charge under curve from TA to T

FT intermediate variable read from F table and used to

calculate F(F = FT + E)

FULCH full recharging capability of Batt I (near 40 for the

first five recharges)

JBATT(I) 0 if Batt I is OFF

l if Batt I is ON

NCH number of times the charger has been used on Batt I

NCHRG(I) number of times the charger has been used on Batt I

ONAH(I) amp-hour status of Batt I at ON-time

ONAHS amp-hour status at ON-time

TMO

ON time

ONTIM(I)

ground elapsed time (g.e.t.) at which the charger was

put on Batt I

PT

present ground elapsed time (g.e.t.)

RE

area under charge rate curve from F to G

SEENA

spacecraft electrical energy network analysis

T

time variable for time axis in charge rate curve

(T = PT - ONT)

TA

time of the last solution relative to the time axis of

the charge rate curve

TD(I)

ground elapsed time (g.e.t.) of the last major discharge

of Batt I

TLD

time of the last discharge

TLS

time of the last solution

TL(I)

ground elapsed time (g.e.t.) of the last solution for

the charger on Batt I

VI

volt-amp characteristic of charger

VOLT(I)

voltage on Batt I

XYl

y-intercept on line segment with slope Xl

XX5

y-intercept of line with slope X2

Xl

slope of line segment from (O, A) to (E, B)

X2

slope of line segment from (E, B) to (F, C)

3.0 BATTERY CHARGER DESCRIPTION

The battery charger is of constant voltage output design. The battery charger is supplied with both a dc voltage source of between 25 to 30 volts and an ac voltage source of 400 Hz three-phase 115 volts to produce a 40-volt output charging EMF. A two-stage differential amplifier, Schmitt trigger, current sensing resistor, and voltage amplifier comprise the battery charger logic network which sets up the initial conditions for the battery charger operation. A switching transistor, current amplifier, sensing resistor, and switching choke complete the basic battery charger circuit. The fundamental circuit logic flow is presented in figure 3-1. When the battery charger is turned on, the first stage of the differential amplifier (comparator) is in the ON mode. The second stage of the comparator would then be in the OFF mode, which in turn sets the first stage of the Schmitt trigger to the ON mode with the second stage off. This circuit configuration produces maximum base drive on the current amplifier which turns the switch transistor on. At this time, the circuitry which permits the charging current to flow into the battery is closed. Charging current flows from the transformer rectifier circuit through the switching transistor, current sensing resistor, and the switching choke. voltage drop across the current sensing resistor increases as the charging current increases. When the voltage drop reaches a specific level, the second stage of the comparator is switched automatically to the ON mode, and the first stage is switched to the OFF mode. voltage amplifier in turn is switched off, which reverses the operation mode of the Schmitt trigger. Because the first stage of the Schmitt trigger is off and the second stage is on, the switching transistor is turned off, which interrupts the charging current path. As a result, the electromagnetic field in the choke collapses, discharges into the battery being charged, then passes through the switching diode, and goes through the sensing resistor. As the field in the choke continues to collapse, the EMF in the choke decreases, and the voltage drop across the sensing resistor decreases until the voltage has again reached a specific low level and the comparator mode of operation is reversed (stage 1 is ON, and stage 2 is OFF). The initial conditions have been reset, and one cycle of battery charging has been completed. As a result of a series-connected inductor in the charging circuit, the output current of the charger remains free from variation except for a small amount through the current sensing resistor.

In summary, the battery charger output current is regulated by the sensing resistor until the battery voltage reaches 37.0 volts. At a battery voltage of 37.0 volts, the voltage sensor circuit becomes unbiased and, in conjunction with the sensing resistor, provides a signal for cycling the battery charger. Battery charging is terminated when the battery voltage reaches 39.8 volts.

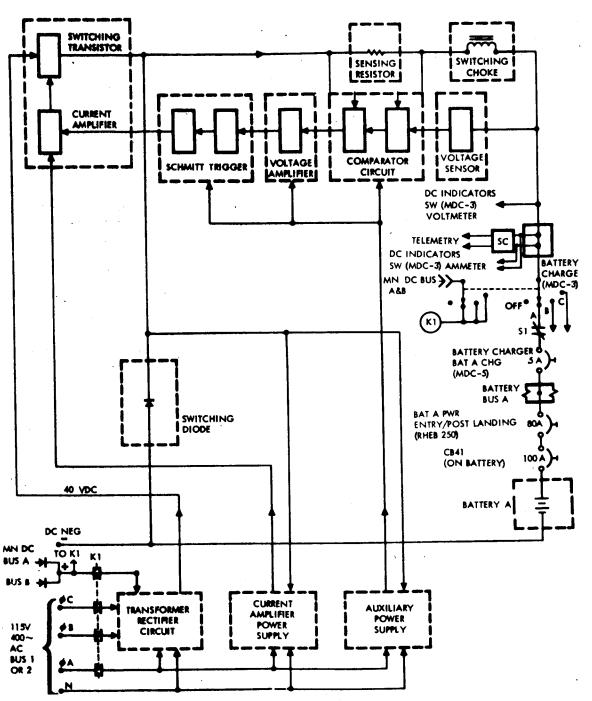


Figure 3-1.- Block diagram of the CSM battery charger circuit.

4.0 ASSUMPTIONS AND DESIGN

The battery charger model is an empirical model. It was designed from actual test and performance data obtained from reference 1. battery charger constraining functions were determined from the actual test data. Various test data trends were noted and are represented in the plots of section 9. A generalized battery charger output curve was determined from the performance data (fig. 4-1). The generalized output curve was broken into three sections so that it could be mathematically manipulated by the initial conditions of the battery being charged. When the initial conditions of the battery are known, the battery charger output characteristic can be calculated. maximum charge time for 100 percent battery recharge can be determined from the charger output characteristic as well as from the battery charger voltage-current characteristics for any time during the recharge cycle of the battery. The battery also can be recharged in a number of charge periods by calculation of the battery charger characteristic output curve for each recharge period.

The battery charger model is based upon several assumptions which are discussed in this section. After the initial inspection of the data in reference 1, it can be determined that the battery charger recharge capability is a function of three variables: the ampere-hours discharged from the battery, the time from the end of the last battery discharge, and the time from the end of the last battery charge. effect of the time from the end of the last battery charge is assumed to be an insignificant factor in the battery charger model. A 100 percent battery charger efficiency is assumed; that is, all of the energy supplied to the battery by the battery charger is used to reenergize the depleted battery. The maximum ampere-hour capacity of the battery is assumed to decrease by 10 percent after five cycles of battery charge and recharge. This decrease simulates battery degradation after several usage cycles. Battery loads of less than 1 ampere per battery will not be considered when the time from the end of the last discharge is being determined. The battery charger input power does not vary as a function of output current in the SEENA CSM program. The AAP CSM battery charger model will have an input power that varies with fluctuations in output charge current. The battery charger input voltage-current levels are indicated in figure 4-2 as a function of the output or charging current.

The general battery charger output or charging curve is shown in figure 4-2. The charger current is plotted against the time the charger is on the battery. All the coordinates except 0.4 can be obtained from the various plots in section 9. The 0.4-ampere coordinate is a test data value which guarantees complete battery recharge. Coordinates A

and B are functions of the ampere-hours discharged from the battery and of the time from the end of the last battery discharge. All the data necessary for these coordinates are given in figures 9-1 and 9-2. Coordinate C is a function of the ampere-hours discharged from the battery and is illustrated in figure 9-3. The data in figure 9-3 are expressed as a factor that is multiplied by the value of coordinate B to obtain coordinate C. Coordinate E is a function of the time from the last discharge. The rise time of the peak current, which is coordinate E, is presented in figure 9-4 as a function of the time from the end of the last discharge. Coordinate F is presented in figure 9-5 as a function of ampere-hours discharged from the battery. The time given on the vertical scale of figure 9-5 is relative to coordinate E. Coordinate G is a function of coordinates A, B, C, E, F, and the total ampere-hours needed to completely recharge the battery.

$G = \frac{2[A-h \text{ needed to completely recharge the battery}] + F}{C + O \cdot 4}$

Because the battery charger output curve can be determined, the effect of the charger on the battery for any period of time can be determined.

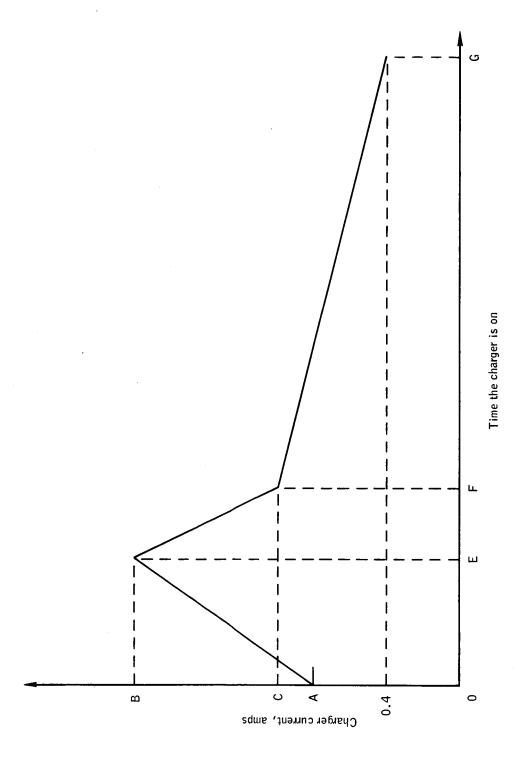


Figure 4-1.- Battery charger general recharge rate curve.

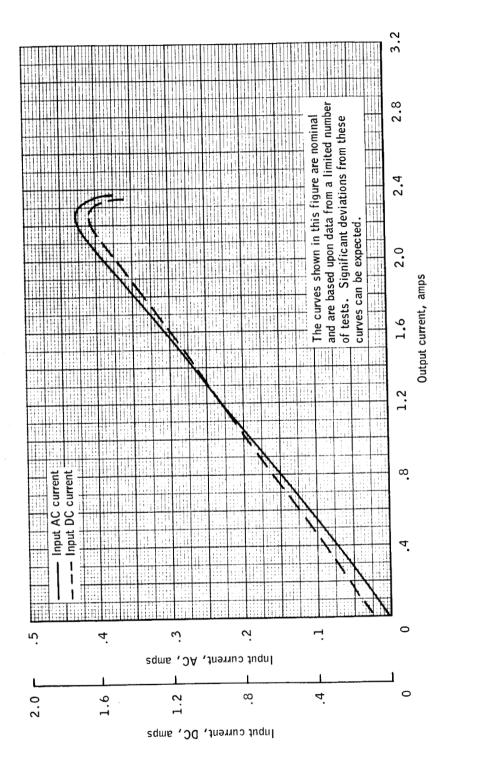


Figure 4-2.- Nominal battery charger input/output characteristics.

30.70

38.31

38.96

38.96

39.16

39.39

Output voltage, V

5.0 COMPUTER PROGRAM DESCRIPTION

The program flow is divided into six major sections in the general flow diagram. These sections emphasize the overall operation of the model from the inputs, through the checks and calculations, and finally to the update and outputs to the main program. Each section will be considered individually in the discussion which follows.

The inputs to the battery charger model consist of three types.

- a. Data curves: these six curves have been extracted from test data on the charger and provide the program with the necessary information to calculate the characteristics of the battery.
- b. Status of the batteries: this vital information is stored in arrays TL, AHS, TD, ONAH, ONTIM.
- c. Charger ON: These data indicate which battery is being charged and at what time a solution point is desired. The form of this input is arranged in a card as follows.

Columns
12 14 16

Present
time
Cols 1-10

Columns
12 14 16

Battery
A B C

"l" if charger is on

The program reads the information in columns 12, 14, and 16 into a three-element array JBATT(3). Then a DO loop is used to compare each element of JBATT to 1. For example, if Batt A is being charged, then JBATT(1) is equal to 1 and control leaves the DO loop; all status information on battery A would be transferred into nonsubscripted names ONT, TLD, TLS, NCH, ONAHS, AHST. At this time, the program checks to see if this is the first time solution during the recharge cycle. If so, a counter that registers the number of charges is incremented. Then the program checks to see how many times battery A has been recharged. If it has been charged more than five times, then an adjustment is made so that the battery can charge back up to only 90 percent of the original charge, which is 36 A-h. This maximum charge is called FULCH in the program.

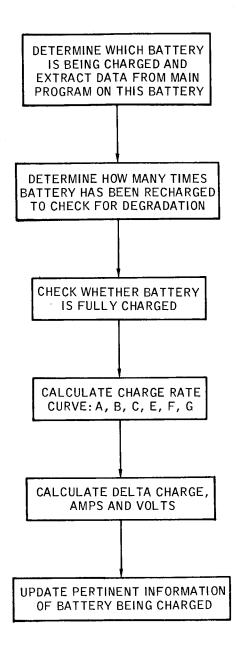
Next, the program checks to see if battery A is fully charged. This check is made by a comparison of AHST + 0.001 to FULCH. The 0.001 is added to insure that the test does not fail in an insignificant decimal place.

Then the program proceeds to the calculation phase. First, the charge rate curve is calculated by interpolation of the curves A, B, C, E, and F which are functions of TLD and AHD. Then by integration from the time of last solution to the present time, the delta charge is calculated. Two checks are made to insure that the charge does not exceed the full charge and that the charger current is 0.4 amps at full charge. The program examines the amount of charge necessary to return to full charge and checks to see that the time to obtain this charge is greater than F. If not, the program will force a solution to come at full charge and 0.4 amps. Otherwise, the manner in which the time G is calculated will insure that the full charge will come at 0.4 amps.

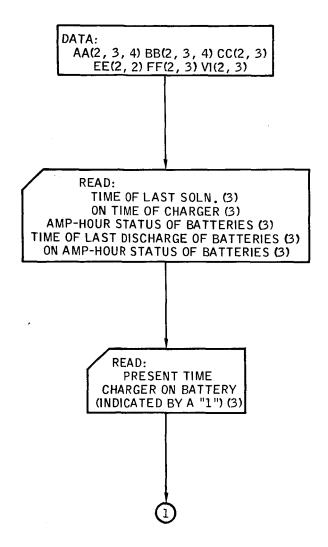
The output voltage is calculated by interpolation of the volt-amp characteristic curve of the charger. Then all pertinent information concerning the battery being charged is updated and stored for usage in the main program.

6.0 FLOWCHART

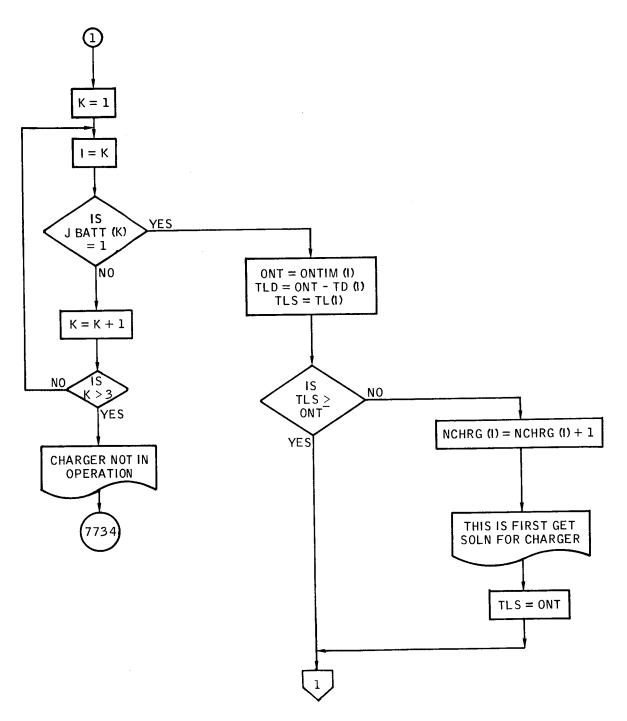
The battery charger program flow chart is presented in this section.



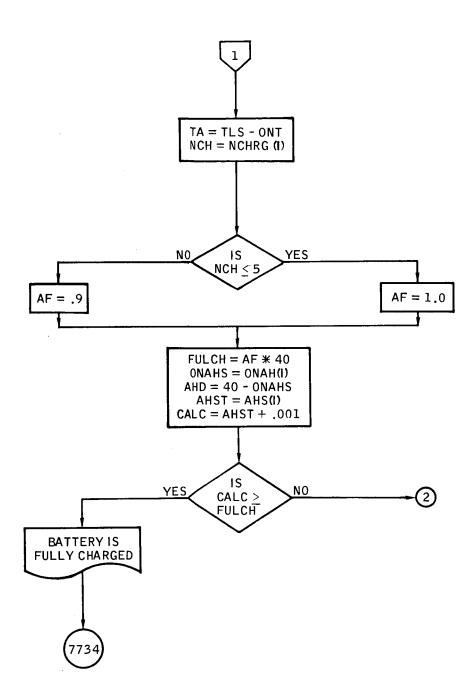
GENERAL FLOW



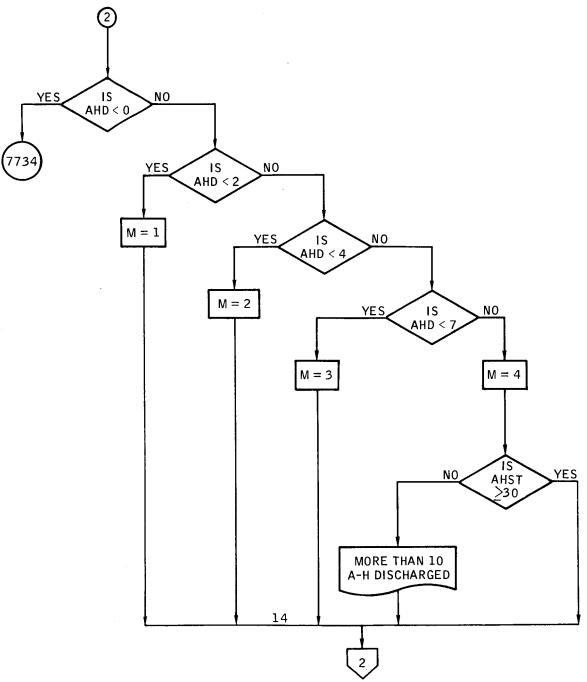
Flow Chart .- Inputs.



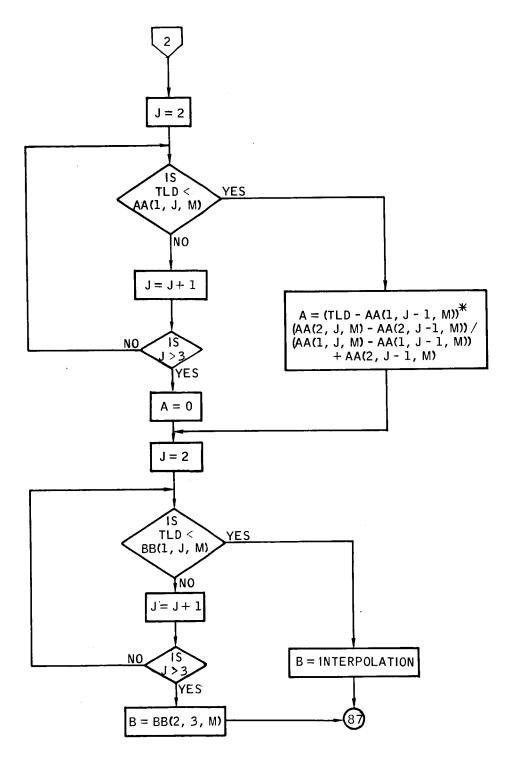
Flow Chart . - Continued.



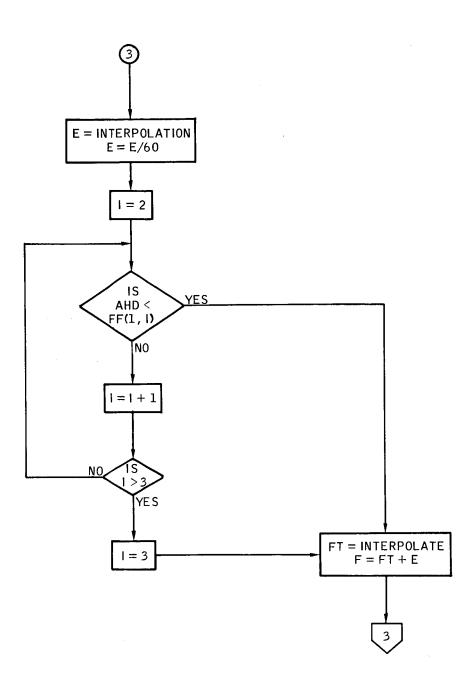
Flow Chart . - Continued.



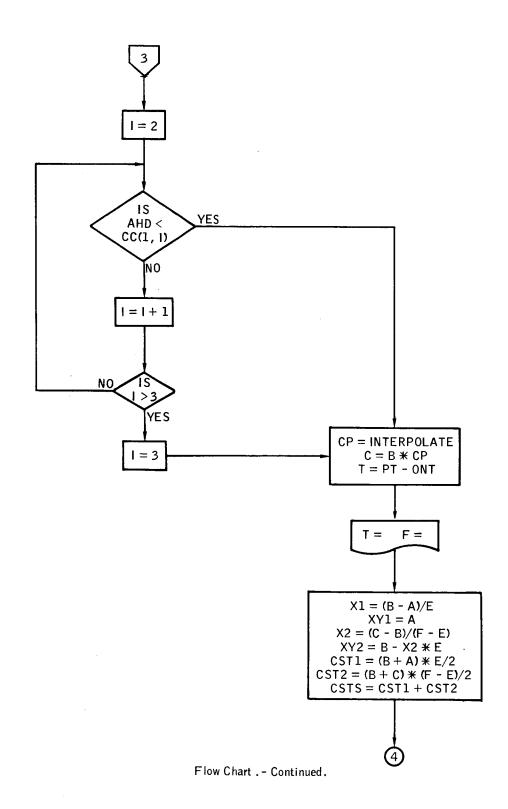
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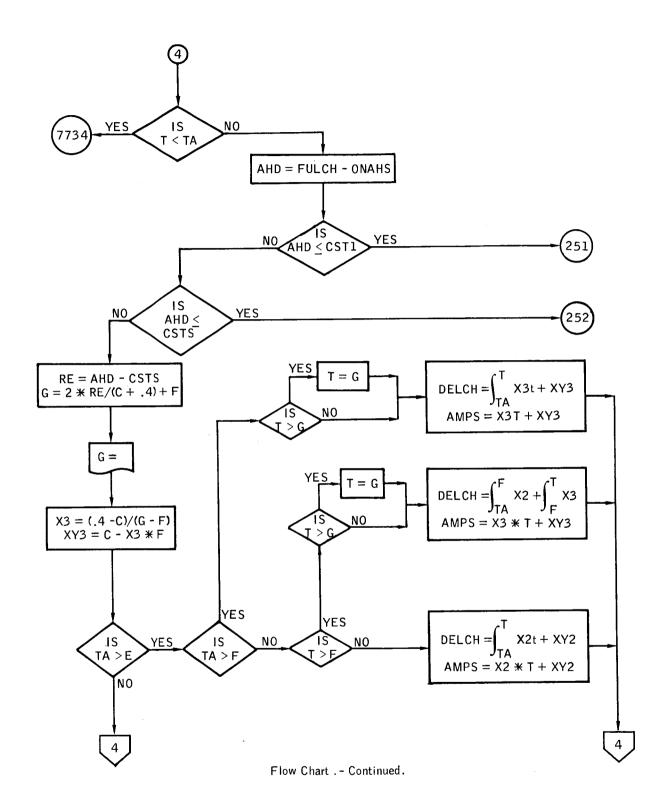


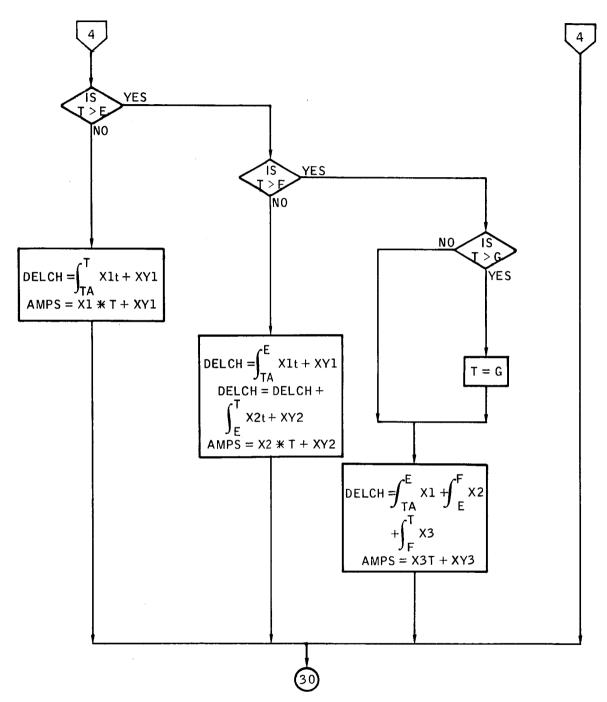
Flow Chart. - Continued.



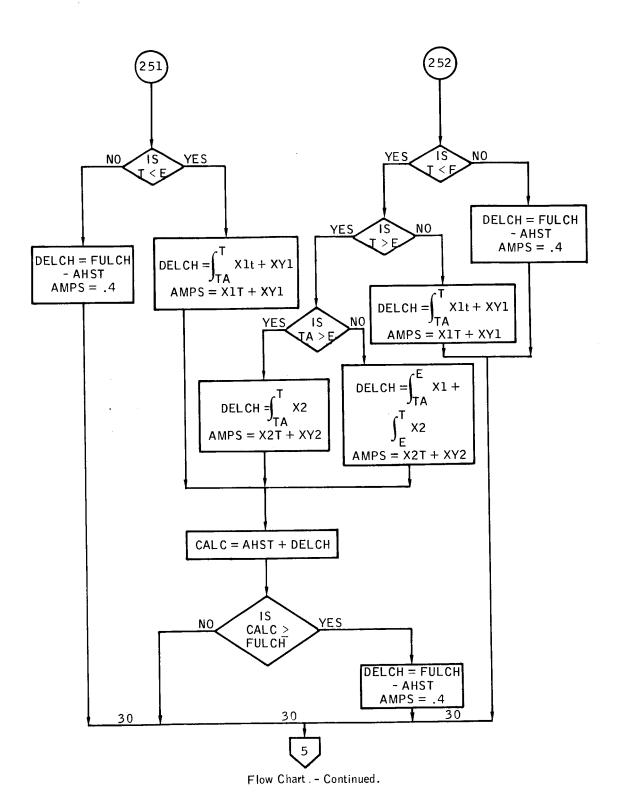
Flow Chart . - Continued.

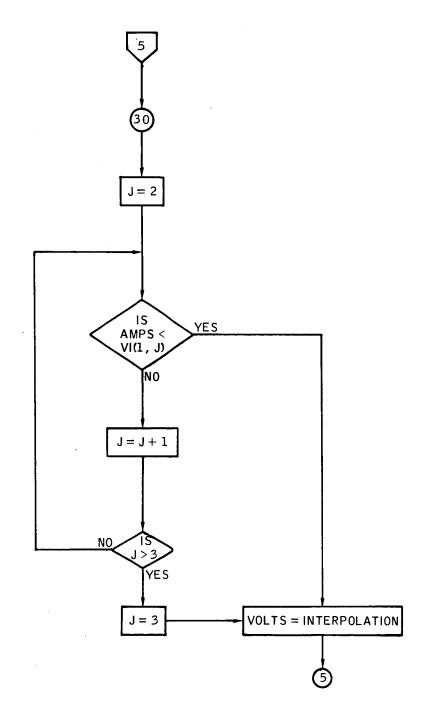




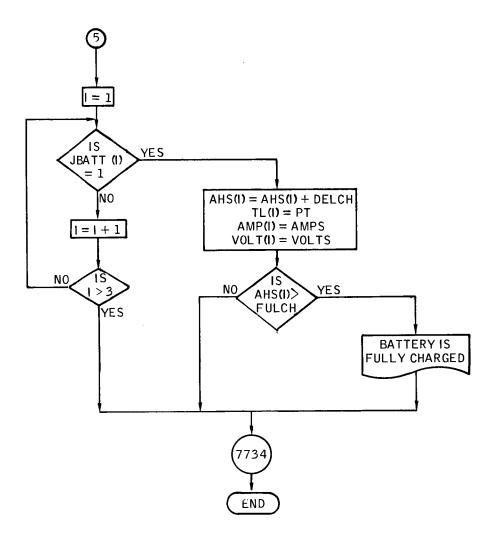


Flow Chart . - Continued.





Flow Chart - Continued.



Flow Chart . - Concluded.

7.0 BATTERY CHARGER MODEL COMPUTER PROGRAM LISTING

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00524	* HU.7	203 FORNIAT (4HDA =:F11-7/1X+3HB =:F11-7/1X+3HC =:F11-7/1X+3HE =:F11-7/
0.0524	405	21A+3HF #+F11+7/1X+7HDELCH #+F11+7/1X+4H1A #+F11+7/1X+4HF; #+F11+7/
00524	2005	31X+9HCF =.F11-7)
600524		11403
00524	209° C	
92500		
00530	2110	1F.JBA11(1)*E4*1160 TO 32
00532	217	31 CONTINUE 10 27 27 27 27 27 27 27 27 27 27 27 27 27
5000	2.2	HJ740 + (1)547 C1
00530	215	1,(1) = PT
00537	216*	Ahrill # Ahrs
00500	217.	
1 4 5 0 0	817	. 1
0.0545	414 220•	34 WKI LEGESS STERY 15 FULLY CHARGED)
9#500		7734 HRITE161661777 T. AMS. TL. ONTIN
00571	222*	MRITE(6,92) AMP VOLT
50900 00604	2230	92 FORMAT(IX,SM ARES,1UA,3F2U-5/14,6M VOLT3,94,3720-5) 64 FURNAT(I3HOPRESENT TIME*F12-5/1/30x+1HA+20x+1HB+2UX+1HC/14+14H LAS
DO 9 0 d	225*	ET GET SOLN-IX 3F ZU-5/1X 11H AT STATUS. 4x 3F 20-5/1X 15H LAST DISCH
0000	756	ا ت
00007	227. • DIAGNEST	77 CONTINUE. 17º THE RETURN STATEMENT IS ILLEGAL IN A MAIN PROGRAM. II MAS CHANGEU TO STUP.
00607	2280	RETURN
01900	229*	END
•	END OF UN	UNIVAC IIUB FORTRAN V COMPILATICM - I .DIAGNOSTIC. MESSAGEIS)

8.0 SAMPLE CASE

The battery charger output characteristic is presented in figure 8-1 as a function of the battery initial conditions. The initial conditions are given in table 8-I. Note from figure 8-1 that it takes 5.92 hours to recharge the CSM entry and postlanding battery which has been depleted by only 4 A-h. This recharge time is caused primarily by the very long time period of 22 hours from the time of last discharge to the time the battery charger was put on the battery. Very low starting and peak charger currents are also results of this long period of time between battery use and battery recharging.

TABLE 8-I.- ENTRY AND POSTLANDING BATTERY

INITIAL CONDITIONS

Conditions tested	Result of test
Battery A-h status at time of charging	 36.0
Full charge A-h rating of battery	 40.0
Time from end of last discharge, hr	 22.0

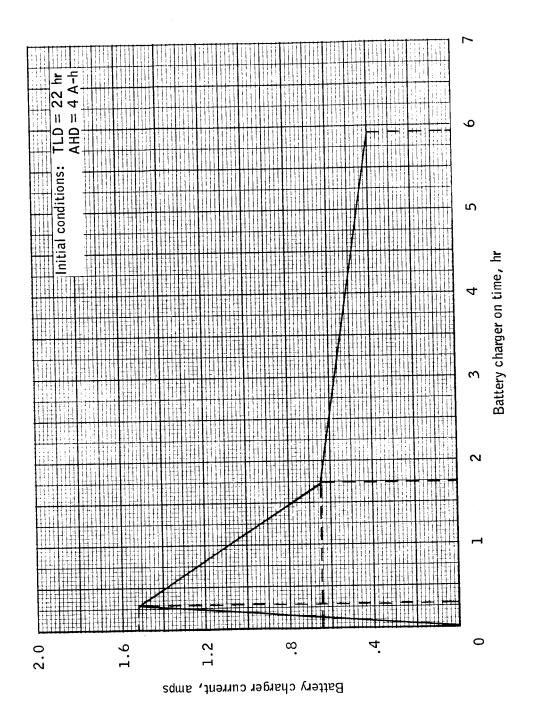


Figure 8-1.- Battery charger output profile.

9.0 BATTERY CHARGER MODEL DESIGN DATA

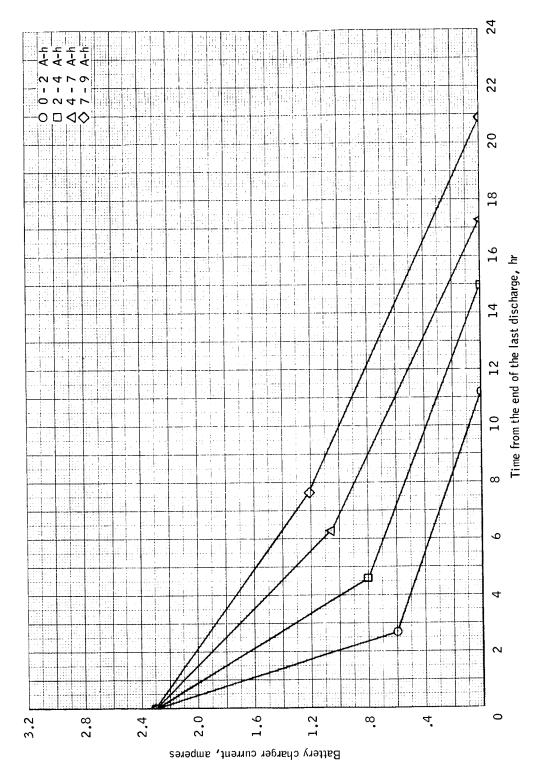


Figure 9-1.- Battery charger starting current co-ordinate A.

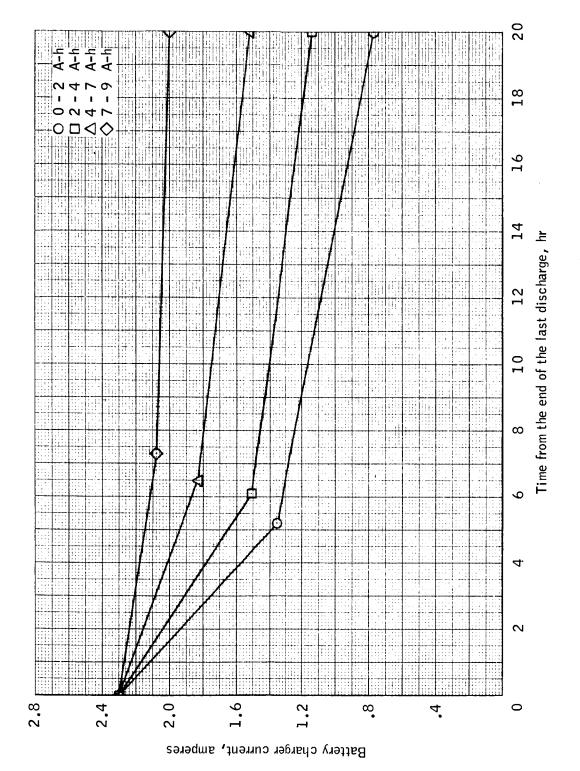


Figure 9-2.- Battery charger peak output current co-ordinate B.

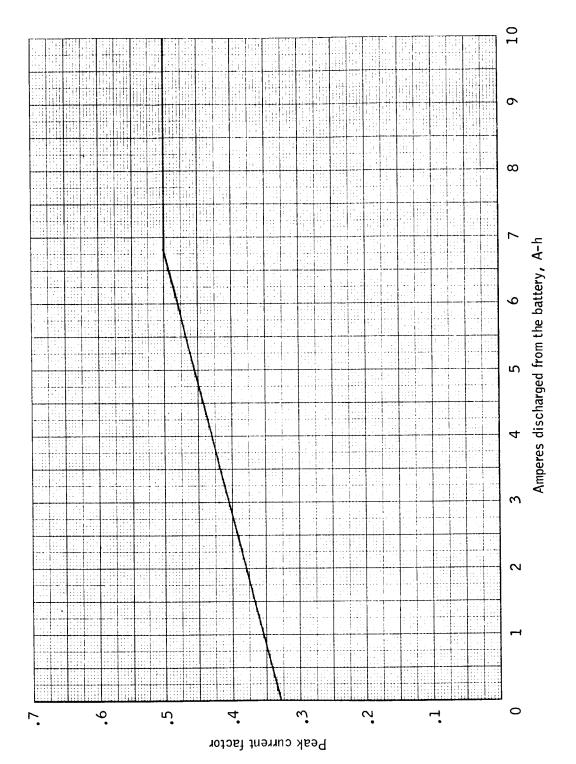


Figure 9-3.- Battery charger output current co-ordinate C peak current factor.

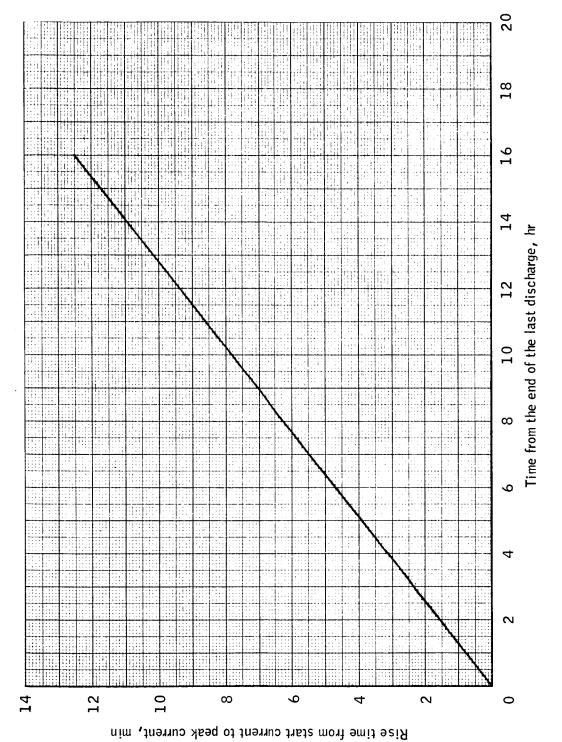


Figure 9-4.- Battery charger output current rise time from the starting current level to the peak current level co-ordinate E_{\bullet} .

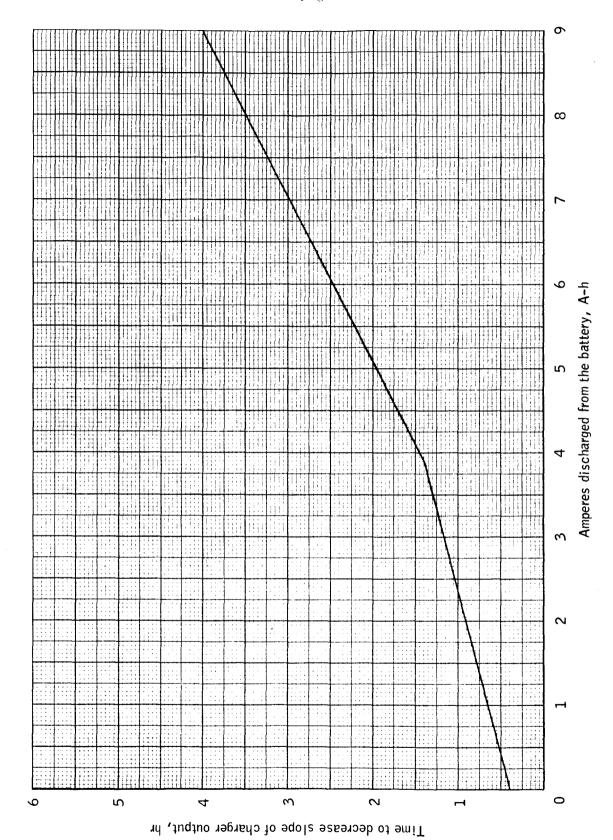


Figure 9-5.- Battery charger output co-ordinate F.

REFERENCES

- 1. Charge Discharge Testing of the Entry and Post-Landing Battery and Spacecraft Battery Charger. MSC/IESD document 21-15, Revision A.
- 2. Spacecraft Operational Data Book, SNA-8-D-027(I) Rev I, Electrical Power Subsystem.